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DESCRIPTIVE SURVEY OF HIGH SPEED FLOW SEPARATION

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and

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FLIGHT DYNAMICS LABORATORY
AIR FORCE WRIGHT AERONAUTICAL LABORATORIES
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FOREWORD

This report was prepared by Reservists assigned to the Aeronautical Systems Division (ASD) for the Air Force Wright Aeronautical Laboratories (AFWAL) located at Wright-Patterson Air Force Base, Ohio. The ASD Reserve Project number was 77-052.

The Air Force Project Monitor was Richard Neumann (AFWAL/FIMG) and the work was performed as part of Air Force Task 240407, "Aeroperformance and Aeroheating Technology". Richard Neumann formulated the project, guided our efforts, and continually supported the needs of the project. The authors thank Richard Neumann and James Hayes (AFWAL/FIMG) for their substantial contributions to this project.

The authors also wish to acknowledge and thank the numerous Air Force Reservists who contributed to this work: LTC Allan Dean, MAJ Paul Beck, MAJ Myron Goldman, MAJ Jack Grimes, MAJ Harold Jensen, MAJ Robert Kirchner, CPT Lawrence Crain and CPT Eric Holwitt.

This memorandum has been reviewed and is approved.

. CHRISTOPHER BOISON

Chief, High Speed Aero Performance Branch

Aeromechanics Division

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Section I

Flow separation can substantially alter the anticipated pressure and heating loads on high speed aircraft surfaces, and can compromise the design of high speed vehicles. The importance of this problem is well recognized and has prompted several hundred scientific investigations of various aspects of separated flow phenomena. Many of these investigations are listed and categorized herein. Investigations of lee side separated flow phenomena, however, are not included.

Three major portions of this report are: 1) the inverse chronologic listing of reports of high speed separated flow investigations, 2) a descriptive table and discussion of the investigations of various aspects of high speed flow phenomena, and 3) a reference list emphasizing the more recent investigations. We have also included a cross reference list of first authors and date of report. These are described in more detail in the following text. This report is useful in describing features of various aspects of high speed flow separation phenomena and delineating recent and appropriate references.

Section II

LITERATURE SEARCHES

Many library searches were accomplished in gathering material pertinent to high speed flow separation. "Key Words" were used to obtain bibliographies from: the U. S. Air Force, the U. S. Navy, NASA, the American Institute of Aeronautics and Astronautics (AIAA), and Grumman Aerospace Corporation. The proliferation of reports prompted our formulating a computer program for listing the reports in an orderly fashion (reverse chronological and alphabetical by author). In addition to the various library searches, reference lists from each report were reviewed to obtain additional references pertinent to high speed flow separation.

Emphasis has been placed on the more recent investigations, although landmark older references are also listed. The computer printout lists the date, year and month, of the reference, and the first 18 letters of the title. Numbers following the date are simply a numerical listing, alphabetically by author of the reports issued in a particular month. To our knowledge, this is the most complete listing of investigations of high speed flow separation phenomena currently available.

Date, Authors, and initial portions of report titles are listed in Table I, "Chronologic List of Reports."

Section III

DESCRIPTION OF TYPES OF SEPARATION AND INVESTIGATIONS

Experimental and theoretical investigations of high speed flow separation phenomena are listed and categorized in Table II. The date is the same as that given in the computer listing (Table I), but the names of the first authors have been reduced to just the first eight letters. Thirty-five categories are used to delineate features of each report. These separated flow phenomena and selected investigations are described below. Only a few investigations are singled out for specific examples but every reference listed was carefully reviewed.

The letters in the table are the same as the first letter of the column heading; an aid in reading the tables. Column headings are described below.

The first four columns describe the general nature of the report. Included under "survey" are reports that include much information gleaned from many other investigations. One of the more outstanding survey reports, because of its thoroughness, is that prepared by Ryan. Of course, most reports in this complex field are experimental in nature. The theoretically oriented reports deal primarily with two-dimensional flows. Many of the theoretical reports lean heavily on numerical analyses and computer codes. Even though a numerical program is operational at one facility, it may require much work to make it operational at another facility.

Two-dimensional and axisymmetric separated flows are somewhat similar in theory. Conical flares or rocket exhaust plumes being the axisymmetric counterpart of two-dimensional ramps, flaps, or external burning. Much theoretical work in this area has been accomplished by Holden. Many have conducted experimental investigations and an empirical data base, along with analytical methods, has been established for predicting pressure and heat transfer distributions. The methods are practically useful for engineering needs, but it should be noted that the steadiness of two-dimensional separated flows is questionable. Indeed, Ginoux shows evidence of three-dimensional flow effects at reattachment of pseudo two-dimensional separated flows.

Three-dimensional separated flows are untenable to purely theoretical methods. There are, however, many numerical methods (such as those of Hankey, Shang et al) and analytical empirical methods (such as those of Neumann and Hayes) for predicting adequately facets of three-dimensional separated flows. Flap aspect ratio and end plate effects are discussed in several reports (Ball, Cassel, Kaufman, Neumann and others).

Hankey, Shang, and others have addressed the corner flow problem, numerically. This is treated as a conical flow and is different from the disturbance caused by a fin mounted on a surface (Korkegi, Neumann and others).

Edwards, Kaufman, Whitehead, and others investigated wing sweep effects on separated flows ahead of trailing edge controls. These effects are most pronounced when there is a change in the character of the boundary layer over the surface (from laminar outboard to turbulent inboard, transition occurring parallel to the swept leading edge).

NASA, the Air Force, other government agencies, and many private corporations have written "white papers" on high speed aircraft and missile design, and incorporation of the vehicle design with the (scramjet) propulsion system.

Chapman, Sterrett, Zukoski, and many others have investigated separated flows ahead of forward facing steps. These flows are now fairly well understood and their characteristics predictable for practical engineering purposes. Separated flows ahead of ramps or flaps pose a far more severe theoretical problem (the reattachment location is unknown initially). Furthermore, there is an unsteadiness in these flows that can cause control "buzz". Nevertheless, there is a wealth of experimental data useful in pinning down salient aspects of the flows.

There are many varieties of "base" flows. This category includes: rearward facing steps, blunt bases of axisymmetric bodies, flows over sharp expansion corners, wake flows and plume induced flow separation (which may also be included under "ramp"). Many base geometries and flow parameters proliferate a numerous variety of flows, precluding inclusive analyses of all such type flows. Of course many investigations provide guidelines, but an experimental approach for a particular vehicle and flow conditions still appears to be mandated.

The desire to release internal stores stably at high speeds has led to many investigations of cavity flows (Heller, Rossiter and others). Two-dimensional cavity flows and cavities on axisymmetric vehicles (applicable to high-altitude deceleration) have been investigated. However, the bulk of high speed cavity flow investigations pertain to flows past bomb bays. The steadiness of the flow and accoustics are important aspects of this problem. Criteria are available for analyzing and even predicting certain facets of cavity flows, but even fundamental scaling laws have yet to be established.

Ericsson, Stetson and others have performed many investigations of flows over forward facing spikes and cavities in nose cones. Again, stability of these flows is of paramount importance. Ablating nose cones are also considered under this heading.

Amick, Kaufman, Voitenko, Werle and others have performed experimental and theoretical investigations of flows past transverse jets. The jet fluid boundary has occasionally been represented as a forward facing step or, three-dimensionally, as a cylinder mounted perpendicularly to the surface.

Flows past protuberances from a surface have much practical significance and have accordingly received much attention. Some investigators that come to mind here are: Couch, Dolling, Gillerlain, Hayes, Kaufman, Korkegi, Lucas, Neumann and Waltrup. The shock wave associated with the protuberance separates the boundary layer from the vehicle surface and results in a complex viscid-inviscid three-dimensional flow field. Neumann and Hayes present practical methods for estimating the extent of separation and the increased pressure and thermal loads on a surface adjacent to sharp leading edge fins. The boundary layer character on the vehicle surface is an important parameter. For blunt fins, or cylinders, not only the character of the boundary layer but also its thickness relative to the fin height and diameter are important parameters.

The shock generated by a fin, or other protuberance, is incident to the boundary layer on the vehicle surface. Other incident shock-wave boundary-layer interactions are created by adjacent bodies, such as the shuttle and the main fuel pod, or stores mounted on wings. The two-dimensional case has been analyzed and "free interaction" theories developed that are reliable. Three-dimensional interaction cases are more complex and frequently require interpolation between sets of experimental data. In our experience, the

extent of the three-dimensional viscid-inviscid interaction flow region is considerably larger than would be anticipated using inviscid flow analyses.

Edney made the definitive investigation of shock impingement. The shock wave ahead of a protuberance separates the boundary layer form the surface. The "dead air" region forms an effective wedge and gives rise to a shock wave. This secondary shock wave impinges on the protuberance shock wave. A slip line starting at the juncture of the shock waves impinges on the protuberance leading edge. The energy of this small portion of the flow field is enormous and results in extremely large heating rates locally at a particular spot on the protuberance leading edge (Fig. 1). Indeed, the destruction of a ventral fin on the X-15 was a result of this type of flow interaction. The interaction also results in local high pressure regions on the surface ahead of the fin. Various types of shock impingement interaction flows can occur. These were categorized by Edney and investigated by him, Keyes, Voitenko, and others.

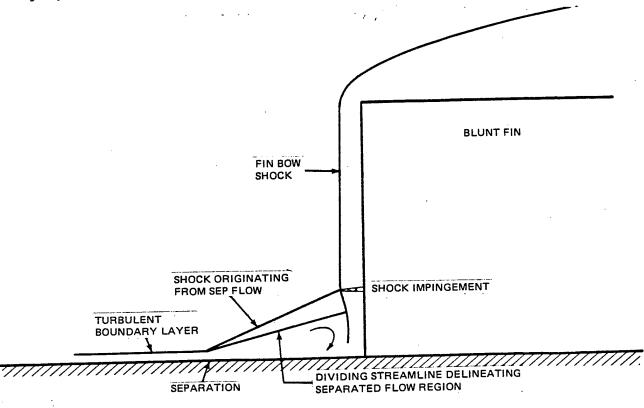


Fig. 1 Centerplane Sketch of Flow Separation and Impingement on a Blunt Leading Edge Fin

As would be expected, boundary layer characteristics are of paramount importance in shock-wave boundary-layer interaction flows. The definitive work here is that of Chapman, though many theoretical and experimental investigations have advanced knowledge in these types of interactions since Chapman's work. Transitional boundary layers are not as readily amenable to theoretical analyses as either laminar or turbulent boundary layers, but are prevalent in wind tunnel experiments. Indeed, the extent of transition is frequently comparable to the extent of laminar boundary layer flow. The lower case "t" used in this column is simply to distinguish it from the adjacent upper case "T" used for turbulent boundary layer interaction flows.

The vast majority of experiments have been conducted using wind tunnels; flight test is considerably more complicated and expensive.

The following columns indicate the type of information presented, whether calculated or measured. Surface static pressure and heat transfer rate distributions are most common. Flow field characteristics, whether calculated or measured, are harder to obtain and therefore appear less frequently in the literature.

The column headed "SCHL-SHD" pertains to all and any type of flow field photograph: schlichting, shadowgraph, interferrograph, holograph, or vapor screen.

Surface oil flow photographs and motion pictures delineate not only the extent of separation but many other features of the interaction flow. These features include the character of the boundary layer (laminar, transitional, or turbulent), the steadiness and stability of the interaction flow, surface streamline directions, and the extent of influence of the interaction flow. Ericsson, Gillerlain, Hayes, Kaufman, Kitchens, Neumann, Sedney and Winkelmann are just a few who have used the surface oil flow technique successfully.

Carriere, Chapman, Gadd, Hung, and several others have formulated expressions for the extent of separation as well as the "free interaction" pressure rise at the onset of separation. These correlations are generally adequate for engineering purposes.

A few reports touch upon other topics, and are usually described more fully under "Notes." These are self-explanatory.

An author-chronologic cross reference list is given in Table III to facillitate recalling the works of specific scientists in this field.

Section IV

CONCLUSIONS

Many investigations of high speed flow separation were reviewed. Salient aspects of interaction flows are discussed. Pertinent references are categorized in a descriptive table.

This reference work was designed to be a useful tool for aerodynamicists interested in the effects of high speed flow separation. At the least, it narrows the number of references to be reviewed in a particular effort involved with a particular aspect of high speed separated flow phenomena.

TABLE I CHRONOLOGIC LIST OF REPORTS

8203		ERICSSON REDING	DYNAMIC SIMULATION THROUGH ANALYTIC
8201	1	HORSTMAN SETTLES W	A REATTACHING FREE SHEAR LAYER IN C
8201	2	SETTLES WILLIAMS B	REATTACHMENT OF A COMPRESSIBLE TURB
8201	3	SHANG HANKEY SMITH	FLOW OSCILLATIONS OF SPIKE-TIPPED B
8111	•	DODS COE	CROSSFLOW EFFECTS ON STEADY AND FLU
8110		TAI	DETERMINATION OF THREE-DIMENSIONAL
8109	1	HUNG CHAUSEE	COMPUTATION OF SUPERSONIC TURBULENT
8109	2	KAUFMAN JOHNSON	METHODS FOR ESTIMATING PRESSURE AND
8108	_	COUSTEIX HOUDEVILL	SINGULARITIES IN THREE-DIMENSIONAL
8105	1	DOLLING BOGDONOFF	SCALING OF INTERACTIONS OF CYLINDER
8105	2	ERICSSON	AEROELASTICITY, INCLUDING DYNAMIC E
8104	1	MEIER GRONAU	VISCOUS AND INTERACTING FLOW FIELD
8104	2	TOBAK PEAKE	TOPOLOGY OF THREE-DIMENSIONAL SEPAR
8103		RIEBE PITTMAN	AERODYNAMIC CHARACTERISTICS OF A HY
8101	1	DOLLING BOGDONOFF	UPSTREAM INFLUENCE SCALING OF SHARP
8101	2	NESTLER	AN EXPERIMENTAL STUDY OF CAVITY FLO
8101	3	NEUMANN HAYES	PROTUBERANCE HEATING AT HIGH MACH N
8101	4	SETTLES PERKINS BO	UPSTREAM INFLUENCE SCALING OF 2D +
8101	5	ZUMWALT	EXPERIMENTS ON THREE-DIMENSIONAL SE
8012		ARDONCEAU ALZARY A	CALCUL DE L'INTERACTION ONDE DE CHO
8011	1	BRANDEIS ROM	THREE-LAYER INTERACTIVE METHOD FOR
8011	2	DAVIS MALCOLM	TRANSONIC SHOCK-WAVE/BOUNDARY-LAYER
8010		BOGDONOFF SETTLES	SEPARATED FLOW AND BOUNDARY LAYER R
8008	1	CASSEL MCMILLEN TA	FINITE SPAN EFFECTS ON FLAP HEATING
8008	2	HANKEY SHANG	ANALYSES OF PRESSURE OSILLATIONS IN
8008	3	HUSSAINI BALDWIN M	ASYMPTOTIC FEATURES OF SHOCK-WAVE B
8008	4	MATEER VIEGAS	MACH AND REYNOLDS NUMBER EFFECTS ON
8007	1	CASSEL JARRETT	HYPERSONIC FLOW OVER SMALL SPAN FLA
8007	2	GOLDSTEIN	WORKSHOP REPORT FOR THE AIAA 5TH AE
8007	3	PEAKE TOBAK	THREE-DIMENSIONAL INTERACTIONS AND
8007	4	SETTLES PERKINS BO	INVESTIGATION OF THREE DIMENSIONAL
8006	1	MACIULAITIS	IMPROVED PREDICTION OF FREQUENCY MO
8006	2	SHILOH SHIVAPRASAD	MEASUREMENTS OF THE TRANSVERSE VELO
8006	3	TIPTON	WEAPON BAY CAVITY NOISE ENVIRONMENT
8004	1	JARRETT CASSEL MCM	FINITE SPAN EFFECTS ON FLAP HEATING
8004	2	SIMPSON CHEW SHIVA	MEASUREMENTS OF A SEPARATING TURBUL
8003	1	JOHNSON KAUFMAN	HIGH-SPEED INTERFERENCE HEATING LOA
8003	2	KUHN	CALCULATION OF SEPARATED TURBULENT
8003	3	PAYNTER	ANALYSIS OF WEAK GLANCING SHOCK/BOU
8003	4	PEAKE TOBAK	THREE-DIMENSIONAL INTERACTIONS AND
8003	5	REDING GUENTHER JE	SCALE EFFECTS ON FLUCTUATING PRESSU
8001	1	CLARK KAUFMAN MACI	AEROACOUSTIC MEASUREMENTS FOR MACH
8001	2	HUNG CLAUSS	THREE-DIMENSONAL PROTUBERANCE INTER
8001	3	MARCONI	SUPERSONIC INVISCID CONICAL CORNER
8001	4	ROSEN PAVISH ANDER	CORRELATION TECHNIQUE FOR PREDICTIN
8001	5	SICLARI	INVESTIGATION OF CROSSFLOW SHOCKS O
7912	1	CEBECI KHALIL WHIT	CALCULATION OF SEFARATED BOUNDARY-L
7912	2	KIRCHNER	ANALYTIC INVESTIGATION OF HYPERSONI UNSTEADY PROPERTIES OF SEPARATED PL
7911	1	DYMENT	COMPUTATION OF THREE DIMENSIONAL TU
7911	2	HORSTMAN HUNG	COULDIGITOR OF THEE BILLEKSTONET IN

7911	3	IKAWA	REAL GAS LAMINAR BOUNDARY-LAYER SEP
7911	4	SHAW	SUPPRESSION OF AERODYNAMICALLY INDU
7911	5	TASSA SANKAR	EFFECT OF SUCTION ON A SHOCK SEPARA
7910	1	CRAWFORD	SOME RECENT DEVELOPMENTS IN THE PRE
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7910	3	KAUFMAN	PRETEST REPORT FOR HEAT TRANSFER EX
7909	1	DELERY	ANALYSIS OF THE SEPARATION DUE TO S
7909	2	REDING	FLUCTUATING PRESSURES ON MILDLY IND
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	4	LEGENDRE	SEPARATION OF A FLOW ALONG A LINE O
7907	1	MODARRESS JOHNSON	INVESTIGATION OF TURBULENT BOUNDARY
7907	2	SHANG HANKEY PETTY	THREE-DIMENSIONAL SUPERSONIC INTERA
7907	3		HYPERSONIC INTERFERENCE FLOW FLIGHT
7906	1	CASSEL DUNCAN LAHT	SEPARATED FLOW PROBLEMS WITHIN THE
7906	2	CHOW	COMMENT ON *WALL SHEAR STRESS MEASU
7906	3	KIRCHNER	CALCULATION OF TIME DEPENDENT FLOWS
7906	4	NASH SCRUGGS	REPLY BY AUTHORS TO R D KIRCHNER
7906	5	ROSE MURTHY	FLOW FIELD INVESTIGATIONS IN CORNER
7906	6	SCHEFERS PFEIFFER	FLOW FIELD INVESTIGATIONS IN COMMEN
7906	7	SCHWEIGER ERHARDT	SHOCK-SHOCK AND SHOCK-BOUNDARY LAYE
7906	8	SETTLES FITZPATRIC	DETAILED STUDY OF ATTACHED AND SEPA
7905	1	ERICSSON ALMROTH B	HYPERSONIC AEROTHERMOELASTIC CHARAC
7905	2	NEUMANN HAYES	AERODYNAMIC HEATING IN THE FIN INTE
7903		KAUFMAN JOHNSON	PRESSURE AND THERMAL DISTRIBUTIONS
7902	1	CLARK	EVAVUATION OF F-111 WEAPON BAY AERO
7902	2	SCIBILIA DUROX	ETUDE DE LA FORMATION D'UN DECOLLEM
7902	3	SHEN	SUPERSONIC FLOW OVER A DEEP CAVITY
7901	1	DOLLING COSAD BOGD	THE SCALING OF 3D BLUNT FIN INDUCED
7901	2	EDITORIAL STAFF	U.S. AIR FORCE RESEARCH AND DEVELOP
7901	3	GILLERLAIN	FIN CONE INTERFERENCE FLOW FIELD
7901	4	HANKEY SHANG	THE NUMERICAL SOLUTION TO PRESSURE
7901	5	HORSTMAN HUNG	COMPUTATION OF THREE DIMENSIONAL TU
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7811	2	ERICSSON	ASYMMETRIC UNSTEADY FLOW IN FORWARD
7811	3	HABERCOM	FLOW REATTACHMENT. A BIBLIOGRAPHY W
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7810	3	VAN DEN BERG (ED)	EUROPEAN RESEARCH PROGRAMME ON VISC
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7803	3	OSKAM VAS BOGDONOF	OBLIQUE SHOCK WAVE/TURBULENT BOUNDA
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THE HYPERSONIC LAMINAR BOUNDARY LAY THREE-DIMENSIONAL SEPARATION (VKI) EFFECT OF SHOCK IMPINGEMENT ON HEAT THEORETICAL DEVELOPMENTS IN SUPERSO EXPERIMENTAL STUDIES OF SHOCK WAVE-LAMINAR AND TURBULENT SEPARATION IN INCIPIENT SEPARATION OF AXIALLY SYM THICK, TWO DIMENSIONAL, TURBULENT B ON LAMINAR BOUNDARY-LAYER SEPARATIO SUPERSONIC, TURBULENT BOUNDARY-LAYE TURBULENT HYPERSONIC VISCOUS INTERA PLATE-INJECTION INTO A SEPARATED SU RESEARCH ON HYPERSONIC AND SUPERSON AFRODYNAMIC INTERFERENCE INDUCED BY SWEEP EFFECTS ON SUPERSONIC SEPARAT ACQUISITION AND REDUCTION OF THIN-S THREE-DIMENSIONAL SEPARATION FOR IN AN APPROXIMATE APPROACH TO BASE FLO A SIMPLE CORRELATION FOR INCIPIENT NEW METHOD FOR SUPERSONIC BOUNDARY-CALCULATION OF TURBULENT BOUNDARY L EUROPEAN RESEARCH PROGRAMME ON HYPE EXPERIMENTAL INVESTIGATION OF HYPER SHOCK WAVE-BOUNDARY LAYER INTERACTI SHOCK IMPINGEMENT CAUSED BY BOUNDAR SIDEWALL BOUNDARY LAYER INFLUENCE O TURBULENT BOUNDARY LAYER SEPARATION HYPERSONIC AIRCRAFT BY 2000 PUSHED FFFFCTS OF BLUNTING AND COOLING ON FLOW AND HEAT TRANSFER DURING THE I EUROPEAN RESEARCH PROGRAMME ON VISC PREDICTION OF THE TURBULENT BOUNDAR INTERFERENCE HEATING DUE TO SHOCK W SUPERSONIC, TURBULENT BOUNDARY LAYE NUMERICAL INVESTIGATION OF UNSTEADY REYNOLDS NUMBER EFFECTS ON THE SHOC A SURVEY OF THE EFFECTS OF SMALL PR A FORWARD FACING STEP STUDY: THE ST EFFECT OF SHOCK IMPINGEMENT ON HEAT ANALYTICAL AND EXPERIMENTAL STUDIES COMPUTER PROGRAMS FOR PREDICTING SU INCIPIENT SEPARATION PRESSURE RISE NASA - CONVAIR ACS SPACE SHUTTLE ST SPACE SHUTTLE ATTITUDE CONTROL SYST THREE DIMENSIONAL SEPARATION CRITER HEATING AND FLOW FIELD STUDIES ON A A SIMPLE MODEL OF NORMAL SHOCK WAVE SHOCK WAVE/TURBULENT BOUNDARY LAYER AEROTHERMODYNAMICS OF THE SPACE SHU HEAT TRANSFER DOWNSTREAM OF ATTACHM

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AUTHOR AUTHOR	GLOTOV L KIRLIN SCHEPERS KEYES KORKEGI SEDNEY K SETTLES KORKEGI OSKAM BARYSHEV	BIRCH RU CHIEN CLARK CZARNECK	MILLER AGARD CARTER CHU YOUN	KUCHEMAN ROSENBAU SPAID VATSA WE KENWORTH	
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AUTHOR	RHUDY LUCERO REDA PAC STRIKE B AEROSPAC	GREEN LAPIN CRESCI R KAUFMAN KLINEBER	WATSON M AMICK GOLDBERG MURPHY ROSE	CASSEL D GINOUX KEYES STEWARTS CARAFOLI	PAGE POLAK KA COUCH NESTLER REDING G	
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AUTHOR	BAUM JOHNSON SHEERAN BALL, KOR	A ID ZU AN OM EY	GOLDMAN HOLDEN LEWIS KU NESTLER STRIKE	WALTRUP WERLE CARRIERE KLINEBER REYHNER	BALL BALL HAN MECKLER NEEDHAM WHITEHEA	
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EMERY BARB	6705	01	EMERY LOLL	6511	01	ERDOS PALL	6206	
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GOLUBINSKY	7608	04	GRANGE KLI	6706	03	GRAY	6608	04
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LAW	7506	10	LAW	7506	11	LAW	7506	12
LAW	7406	03	LAW	7401	04	LAW	7307	03
LAWING	7805	02	LAWING	7506	13	LAWRENCE W	6806	01
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MACIULAITI	8006	01	MAGER	5602	-	MAGNAN SPU	6604	02
MAISE ROSS	7408	05	MARCONI	8001	03	MARKARIAN	6811	03
MATEER VIE	8008	04	MAUK	7908		MAULL	6008	
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MEIER GRON	8104	01	MEYER	6807	02	MIKESELL	6608	08
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MURPHY	6910	03	MURTHY ROS	7807	03	NAGEL BECK	7301	06
NAGEL FITZ	6608	09	NAGEL SAVA	6608	10	NARASIMHA	6711	03
NASH SCRUG	7906	04	NEEDHAM	6712	04	NEEDHAM ST	6606	02
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VOITENKO Z	6701	07	VON KARMAN	7402	06	WAGNER CAM	7103	04
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